The paragraph appearing at page 1, after the Title:

This application is a ~~continuation~~Continuation-in-Part of, and claims priority under 35 U.S.C. § 120 to, International patent application number PCT/IB02/03862, filed 19 September 2002, and claims priority under 35 U.S.C. § 119 to Swiss patent application number 2001 1766/01, filed 25 September 2001, the entireties of both of which are incorporated by reference herein.

Insert a new paragraph at page 9, line 32:

Fig. 1a illustrates a schematic view of a portion of a turbine stage of an axial turbomachine having at least one blade row and at least one guide vane row.

The paragraph appearing at page 9, lines 33-35:

Fig. 1~~a,~~ b, c show a diagrammatic excerpt from a cross section of two inner shrouds, opposite to one another, of two blade/vane roots,

Insert a new paragraph at page 10, between lines 9 and 10:

Fig. 1a shows an extract of an axial turbomachine in which at least one rotor blade row 11 and at least one guide vane row 12 is provided. Both rows 11, 12 are separated axially from each other along an axis A of the axial turbomachine. The turbomachine provides rotor blades 13 and guide vanes 14 arranged in at least one of the rotor blade rows11 and guide vane rows 12 and have respective blade roots 2 or vane roots 3 which protrude into fastening contours 10 within the rotor arrangement 1 and stator housing arrangement 1'. A sealing element 4 of plastically deformable material is provided between a blade root 2 or vane root 3 or guide vane row and

rotary flow machine component 15, like an intermediate piece or a heat insulation segment, directly adjoining the blade root 2 or vane root 3.

The paragraphs appearing at page 10, line 10 through page 11, line 13:

Fig. 1~~a~~b represents a partial cross-sectional representation through two immediately adjacent opposite platforms 21, 31 of two blade~~/vane~~ roots 2, 3, which extend in the peripheral direction (see arrow) on a rotor/stator arrangement 1 and which protrude for fastening purposes into the arrangement 1.

Fig. ~~1a~~ 1b shows the cold condition, i.e., the condition of the blade~~/vane~~ roots 2, 3 before the commissioning of the rotary flow machine, which represents, for example, a compressor unit or a gas turbine stage. A layer-shaped sealing element 4 consisting of plastically deformable material is respectively provided on the two flanks 22, 32 directly opposite to one another of the platforms 21, 31, e.g., by a brazed, soldered, or bonded connection C. These sealing elements 4 jointly enclose a cold gap 5 with a cold gap width s_c. The cold gap width s_c has, typically, a distance apart of between 0.01 and 5 mm.

Fig. 1~~b~~c shows the same arrangement in the hot condition, i.e., after the thermal expansion of the two opposite blade~~/vane~~ roots 2, 3 with the platforms 21, 31 has already taken place. The two sealing elements 4 are joined to one another under the action of forces and are at least partially plastically deformed because of the joining forces which are present and by means of which their effective material thickness has been reduced. At the edge regions of the two plastically deformed layers 4 of Fig. 1, lateral squeeze regions 41 have formed which, because of the plastic deformation, also remain after return to the cold condition.

Due to the provision of plastically deformable materials, according to the invention, between two blade~~/vane~~ roots immediately adjacent to one another, preferably between the adjacent platforms

21, 31 of the two blade~~/vane~~ roots 2, 3, an optimum minimum hot gap 6 forms in the hot condition. This has a gap width s_w which, in the best case, is close to zero and is, in any event, substantially smaller than the cold gap s_c.

The paragraphs appearing on page 13:

List of designations

1	Rotor arrangement
1'	Stator housing arrangement
2, 3	Blade/vane root
21, 21', 31, 31'	Platform
22, 32	Side flanks
4	Plastically deformable material, sealing element
41	Squeeze region
42	Wedge end
5	Sealing gap (cold gap)
6	Sealing gap (hot gap)
7, 8	Platform
71, 81	Side flanks of platform 7,8
72, 82	Cooling ducts
73, 83	Side flanks
74	Sealing protrusion
9	Hot gas duct
10	Fastening contour
11	Rotor blade row
12	Guide vane row
13	Blade

14	Guide vanes
15	Machine component
C	Connection